



[4910-13-P]

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2014-0236; Directorate Identifier 2013-NM-184-AD; Amendment 39-17937; AD 2014-16-13]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for all Airbus Model A300 series airplanes. This AD was prompted by our determination of the need to incorporate new life limits for the main landing gear (MLG) barrel assembly, retraction actuator assembly linkage, and flange duct. This AD requires revising the maintenance or inspection program, as applicable, to include the new life limits. We are issuing this AD to prevent reduced structural integrity of the airplane and possible loss of controllability of the airplane.

DATES: This AD becomes effective [INSERT DATE 35 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov/#!docketDetail;D=FAA-2014-0236> or in person at the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC.

For service information identified in this AD, contact Airbus SAS, Airworthiness Office – EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM 116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-2125; fax 425-227-1149.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to all Airbus Model A300 series airplanes. The NPRM published in the Federal Register on April 17, 2014 (79 FR 21651).

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Community, has issued EASA Airworthiness Directive 2013-0210, dated September 11, 2013 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition all Airbus Model A300 series airplanes. The MCAI states:

Some life limits previously defined in Revision 00 of A300 ALS [airworthiness limitations section] Part 1 have been removed [from] that document at Revision 01 and should normally be included in an ALS Part 4.

At this time, there are no plans to issue an ALS Part 4 for A300 aeroplanes.

Nevertheless, failure to comply with these life limits could result in an unsafe condition.

For the reasons described above, it has been decided to require the application of these life limits through a separate [EASA] AD. Consequently, this [EASA] AD requires application of life limits applicable to Main Landing Gear (MLG) barrel assembly, retraction actuator assembly linkage assembly and flanged duct which were previously contained in Airbus ALS Part 1 Revision 00.

EASA AD 2007-0293 [which corresponds with FAA AD 2009-18-15, Amendment 39-16011 (74 FR 48143, September 22, 2009)], which required compliance with the actions specified in ALS Part 1, will be superseded by a new [EASA] AD, requiring compliance with ALS Part 1 at Revision 1.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov/#!documentDetail;D=FAA-2014-0236-0002>.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (79 FR 21651, April 17, 2014) or on the determination of the cost to the public.

“Contacting the Manufacturer” Paragraph in this AD

Since late 2006, we have included a standard paragraph titled “Airworthy Product” in all MCAI ADs in which the FAA develops an AD based on a foreign authority’s AD.

We have become aware that some operators have misunderstood or misinterpreted the Airworthy Product paragraph to allow the owner/operator to use messages provided by the manufacturer as approval of deviations during the accomplishment of an AD-mandated action. The Airworthy Product paragraph does not approve messages or other information provided by the manufacturer for deviations to the requirements of the AD-mandated actions. The Airworthy Product paragraph only addresses the requirement to contact the manufacturer for corrective actions for the identified unsafe condition and does not cover deviations from other AD requirements. However, deviations to AD-required actions are addressed in 14 CFR 39.17, and anyone may request the approval for an alternative method of compliance to the AD-required actions using the procedures found in 14 CFR 39.19.

To address this misunderstanding and misinterpretation of the Airworthy Product paragraph, we have changed the paragraph and retitled it “Contacting the Manufacturer.” This paragraph now clarifies that for any requirement in this AD to obtain corrective actions from a manufacturer, the actions must be accomplished using a method approved

by the FAA, the European Aviation Safety Agency (EASA), or Airbus's EASA Design Organization Approval (DOA).

The Contacting the Manufacturer paragraph also clarifies that, if approved by the DOA, the approval must include the DOA-authorized signature. The DOA signature indicates that the data and information contained in the document are EASA-approved, which is also FAA-approved. Messages and other information provided by the manufacturer that do not contain the DOA-authorized signature approval are not EASA-approved, unless EASA directly approves the manufacturer's message or other information.

This clarification does not remove flexibility previously afforded by the Airworthy Product paragraph. Consistent with long-standing FAA policy, such flexibility was never intended for required actions. This is also consistent with the recommendation of the Airworthiness Directive Implementation Aviation Rulemaking Committee to increase flexibility in complying with ADs by identifying those actions in manufacturers' service instructions that are "Required for Compliance" with ADs. We continue to work with manufacturers to implement this recommendation. But once we determine that an action is required, any deviation from the requirement must be approved as an alternative method of compliance.

We also have decided not to include a generic reference to either the "delegated agent" or "design approval holder (DAH) with State of Design Authority design organization approval," but instead we have provided the specific delegation approval granted by the State of Design Authority for the DAH throughout this AD.

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting this AD with the changes described previously and minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (79 FR 21651, April 17, 2014) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM (79 FR 21651, April 17, 2014).

We also determined that these changes will not increase the economic burden on any operator or increase the scope of this AD.

Costs of Compliance

We estimate that this AD affects 7 airplanes of U.S. registry.

We also estimate that it takes about 1 work-hour per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work hour. Based on these figures, we estimate the cost of this AD on U.S. operators to be \$595, or \$85 per product.

Authority for this Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds

necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov/#!docketDetail;D=FAA-2014-0236>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800-647-5527) is in the ADDRESSES section.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39 - AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2014-16-13 Airbus: Amendment 39-17937. Docket No. FAA-2014-0236; Directorate Identifier 2013-NM-184-AD.

(a) Effective Date

This AD becomes effective [INSERT DATE 35 DAYS AFTER DATE OF PUBLICATION IN THE FEDERAL REGISTER].

(b) Affected ADs

None.

(c) Applicability

This AD applies to all Airbus Model A300 B2-1A, B2-1C, B2K-3C, B2-203, B4-2C, B4-103, and B4-203 airplanes, certificated in any category.

(d) Subject

Air Transport Association (ATA) of America Code 32, Landing Gear; and 36, Pneumatic.

(e) Reason

This AD was prompted by our determination of the need to incorporate new life limits for the main landing gear (MLG) barrel assembly, retraction actuator assembly linkage, and flange duct. We are issuing this AD to prevent reduced structural integrity of the airplane and possible loss of controllability of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Revise the Maintenance or Inspection Program

Within 90 days after the effective date of this AD, revise the maintenance or inspection program, as applicable, to incorporate the life limits specified in Appendix 1 of this AD into the Airbus A300 Airworthiness Limitations Section (ALS) Part 1. The initial compliance time for the replacement is identified in Appendix 1 of this AD and is prior to the applicable life limits specified in Appendix 1 of this AD, or within 90 days after the effective date of this AD, whichever occurs later.

(h) No Alternative Actions and Intervals

After accomplishing the revision required by paragraph (g) of this AD, no alternative actions (e.g., inspections) or intervals may be used unless the actions or intervals are approved as an alternative method of compliance (AMOC) in accordance with the procedures specified in paragraph (i)(1) of this AD.

(i) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local

Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-2125; fax 425-227-1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Contacting the Manufacturer: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

(j) Related Information

Refer to Mandatory Continuing Airworthiness Information (MCAI) European Aviation Safety Agency Airworthiness Directive 2013-0210, dated September 11, 2013, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov/#!documentDetail;D=FAA-2014-0236-0002>.

(k) Material Incorporated by Reference

None.

Appendix 1 to this AD – New Life Limits for the Main Landing Gear Barrel Assembly, Retraction Actuator Assembly Linkage, and Flange Duct

(*) Whichever occurs first.

Notes are located under the assy title.

		LIFE LIMITS (*)			LIFE LIMITS APPLICABILITY					
	Part Number	FH	LDG	Cal.	B2-1A B2-1C	B2K-3C B2-20x	B2-320	B4-2C B4-1xx	B4-2xx	C4-203 F4-203
ATA 32-10-00 MAIN LANDING GEAR										
BARREL ASSEMBLY										
Stirrup	C66277-10	N/A	66600	N/A			X	X	X	X
	C66277-12	N/A	76600	N/A			X	X	X	X
	C66277-14	N/A	76600	N/A			X	X	X	X
	D58303-1	N/A	76600	N/A			X	X	X	X
Stirrup pin	C66457	N/A	76600	N/A			X	X	X	X
	D48939	N/A	76600	N/A			X	X	X	X
	D48939-1	N/A	76600	N/A			X	X	X	X
	D58314-1	N/A	76600	N/A			X	X	X	X
Universal joint	C66279	N/A	76600	N/A			X	X	X	X
	C66279-2	N/A	76600	N/A			X	X	X	X
	C66279-6	N/A	76600	N/A			X	X	X	X
	D58313-1	N/A	76600	N/A			X	X	X	X
Plate (Upper end)	C61637-10	N/A	76600	N/A	X	X				
	C61637-11	N/A	76600	N/A	X	X				
	C61637-12	N/A	76600	N/A	X	X				
Plate (Rear head end)	C61638-10	N/A	53300	N/A	X	X				
	C61638-11	N/A	53300	N/A	X	X				
	C61638-20	N/A	76600	N/A	X	X				
Tie rod	C68523-3	N/A	76600	N/A	X	X				

**Appendix 1 to this AD – New Life Limits for the Main Landing Gear Barrel Assembly, Retraction Actuator Assembly Linkage,
and Flange Duct (continued)**

(*) Whichever occurs first.
Notes are located under the assy title.

		LIFE LIMITS (*)			LIFE LIMITS APPLICABILITY					
	Part Number	FH	LDG	Cal.	B2-1A B2-1C	B2K-3C B2-20x	B2-320	B4-2C B4-1xx	B4-2xx	C4-203 F4-203
RETRACTION ACTUATOR ASSEMBLY										
(1) When SB A300-32-0123 embodied before SB A300-32-0113.										
(2) When SB A300-32-0123 embodied after SB A300-32-0113.										
Sliding rod	C69028-1	N/A	34 000	N/A	X	X				
	C69028-4	N/A	34 000	N/A	X	X				
	C69029-1 (1)	N/A	32 000	N/A			X	X	X	X
	C69029-2	N/A	32 000	N/A			X	X	X	X
	C69029-3	N/A	32 000	N/A			X	X	X	X
	C69029-4 (2)	N/A	22 000	N/A			X	X	X	X
Piston	C67078	N/A	33 000	N/A			X	X	X	X
	C67078-1	N/A	33 000	N/A			X	X	X	X
End fitting	C61342-4	N/A	36 700	N/A	X	X				
	C66510-4	N/A	32 000	N/A			X	X	X	X
LINKAGE ASSEMBLY										
Upper multiple link pin (Multiple link/Upper link)	C61505	N/A	76 600	N/A	X	X				
	C61505-1	N/A	76 600	N/A	X	X				
	C61505-20	N/A	76 600	N/A	X	X				
ATA 36-11-05 PNEUMATIC										
(1) "xx" at the end of the P/N stands for any number between 00 and 99.										
Duct flanged (1)	A21274063000xx	N/A	24 000	N/A	X		X	X		

Issued in Renton, Washington, on August 4, 2014.

Jeffrey E. Duven,
Manager,
Transport Airplane Directorate,
Aircraft Certification Service.

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